IN THE CLAIMS

1-12. (canceled)

13. (currently amended) A gas turbine engine ~~comprising~~ assembly comprising:

a gas turbine engine; and

a thrust link comprising a first connector, a second connector, and a span section extending therebetween, ~~each said thrust link~~said first connector comprising a first side, a second side, and an opening extending therebetween and defined by an inner surface, said inner surface comprising a plurality of threads extending radially outward into said connector opening such that said threads are axially spaced substantially constant along said inner surface, said plurality of threads comprises a first thread portion positioned proximate said first side and initially engaging a threaded apparatus coupled to said first connector, said plurality of threads comprising a ~~cutback configured to reduce stresses induced into said plurality of threads~~cutback, wherein said cutback extends through said first thread ~~portion.~~portion, said first connector coupled to said turbine engine, said cutback configured to facilitate reducing stresses induced into said plurality of threads.

14. (currently amended) A gas turbine engine assembly in accordance with Claim 13 wherein ~~each~~ said thrust link connector opening has a center axis of symmetry extending from said connector first side to said connector second side, said cutback oblique with respect to said center axis of symmetry.

15. (currently amended) A gas turbine engine assembly in accordance with Claim 14 wherein each said thread comprises a crest and a root, said cutback extending through said first thread portion crest.

16. (currently amended) A gas turbine engine assembly in accordance with Claim 15 wherein said thrust link connector first thread portion is positioned adjacent at least one of said connector first side and said connector second side.

17. (currently amended) A gas turbine engine assembly in accordance with Claim 15 wherein said thrust link connector cutback comprises a first end and a second end, said cutback first end extending through said thrust link connector first thread portion, said cutback second end between said connector first thread portion and said thrust link connector second ~~sides.~~side.

18. (currently amended) A gas turbine engine assembly in accordance with Claim 17 wherein said thrust link connector cutback first end is positioned a first distance from said thrust link connector opening center axis of symmetry, said thrust link connector cutback second end is positioned a second distance from said thrust link connector opening center axis of symmetry.

19. (currently amended) A gas turbine engine assembly in accordance with Claim 18 wherein said thrust link connector cutback first distance is longer than said thrust link connector cutback second distance.

20. (currently amended) A gas turbine engine assembly in accordance with Claim 13 wherein said thrust link connector cutback is further configured to reduce a stiffness factor of at least a portion of said plurality of threads.